

IN THE CLAIMS

Please amend the claims as follows:

Claim 1 (Currently Amended): A turbine nozzle, comprising:
airfoils stacked along ~~the~~a stacking axis,
wherein ~~the~~high curvature portions on a suction surface in ~~airfoil~~section~~sections~~
successively formed along the stacking axis ~~of the airfoil~~ describe a parabola line that curves
toward ~~the~~a pressure side of the airfoil when seen from ~~the~~a front or a rear of the turbine
nozzle, wherein ~~the~~a point of maximum ~~high~~ curvature in each airfoil is located at a
midpoint portions on ~~suction~~ surface in ~~airfoil~~section curve ~~most~~ at the center along the
stacking axis of ~~the airfoil~~ from a straight line that connects with a first intersection between
the parabola line and an inner band in the turbine nozzle and with a second intersection
between the parabola line and an outer band in the turbine nozzle, and wherein a distance of
the point of the maximum curvature falls to the straight line is within a range from 0.02 to
0.03-fold2 to 3 % of a distance between the first and second intersections along the stacking
axis of the airfoil.

Claim 2 (New): A turbine nozzle, comprising:

an inner band;
an outer band; and
a plurality of airfoils disposed between the inner and outer bands, each airfoil of the
plurality having a convex pressure side, a concave suction side, an X axis along a stacking
direction, and a Y axis perpendicular to the X axis, the X axis being defined by a straight line
connecting first and second points at intersections of the airfoil with the inner and outer
bands, respectively, the first and second points defining portions of maximum airfoil

curvature on the concave suction side of airfoil cross sections perpendicular to the X axis, and the Y axis intersecting the X axis at a midpoint between the first and second points, wherein concave suction side profiles defined by cross sections parallel to X-Y planes are parabolic when seen from a front or a rear of the turbine nozzle, and

$$0.02H \leq Y_{\max} \leq 0.03H,$$

where Y_{\max} is the Y coordinate of the midpoint and H is the distance along the X axis between the first and second points.

Claim 3 (New): The turbine nozzle according to claim 2, wherein an airfoil reduced velocity ratio for a lower-degree vibration associated with torsion is 0.7 or less.

Claim 4 (New): The turbine nozzle according to claim 3, wherein a ratio of a natural frequency of the airfoil to a natural frequency of an airfoil without curvature is greater than 1.0.

Claim 5 (New): A low pressure turbine of a gas turbine comprising the turbine nozzle according to Claim 2.

Claim 6 (New): A high pressure turbine of a gas turbine comprising the turbine nozzle according to Claim 2.

Claim 7 (New): A gas turbine engine comprising the turbine nozzle according to Claim 2.

Claim 8 (New): The turbine nozzle according to claim 1, wherein an airfoil reduced velocity ratio for a lower-degree vibration associated with torsion is 0.7 or less.

Claim 9 (New): The turbine nozzle according to claim 8, wherein a ratio of a natural frequency of the airfoil to a natural frequency of an airfoil without curvature is greater than 1.0.

Claim 10 (New): The gas turbine nozzle according to claim 1, wherein a trailing edge of each airfoil is parabolic.

Claim 11 (New): The gas turbine nozzle according to claim 2, wherein a trailing edge of each airfoil is parabolic.

Claim 12 (New): A low pressure turbine of a gas turbine comprising the turbine nozzle according to Claim 1.

Claim 13 (New): A high pressure turbine of a gas turbine comprising the turbine nozzle according to Claim 1.

Claim 14 (New): A gas turbine engine comprising the turbine nozzle according to Claim 1.

Claim 15 (New): A gas turbine engine having a turbine nozzle with redesigned vanes, the turbine nozzle being according to Claim 1.

Claim 16 (New): A gas turbine engine having a turbine nozzle with redesigned vanes, the turbine nozzle being according to Claim 2.